

METHODS AND APPARATUS FOR REDUCING
FLOW ACROSS COMPRESSOR AIRFOIL TIPS

ABSTRACT OF THE DISCLOSURE

A method enables a rotor blade for a gas turbine engine to be fabricated. The method comprises forming an airfoil including a first side wall and a second side wall that each extend in radial span between an airfoil root and an airfoil tip, and wherein the first and second side walls are connected at a leading edge and at a trailing edge, and forming a winglet that extends outwardly from at least one of the airfoil first side wall and the airfoil second side wall, such that a radius extends between the winglet and at least one of the airfoil first side wall and the second side wall.